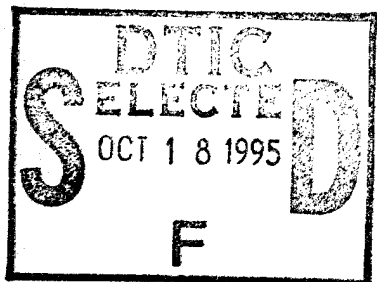


ARMY RESEARCH LABORATORY



Application of Pulsed Flame Velocimetry to the Study of Inhibited Methane/Oxygen Flames



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13. ABSTRACT (Maximum 200 words) Knowledge of flame velocities is an important factor in the design and construction of many devices involving combustion phenomena. We report here results of flame velocity measurement obtained using pulsed flame velocimetry, a new technique that may be useful for Halon replacement research. The method developed here is based on and is an extension of the pulsed flame photometer detector developed at Tel Aviv University. Flame velocities for burning methane/oxygen mixtures to which a small amount of inhibitor has been added are determined using pulsed flame velocimetry. Results are compared to flame velocities measured for similar mixtures using other techniques. Relative flame velocities for burning methane/oxygen mixtures doped with small amounts of inhibitors are compared with flame inhibition efficiencies. The utility of flame velocities determined using pulsed flame velocimetry as a predictive technique for inhibitor efficiency studies is discussed.					
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1. INTRODUCTION

One of the great successes in firefighting technology over the past several decades has been the development and use of Halon 1301 (CF_3Br) and Halon 1211 (CF_2ClBr) as fire-extinguishing agents. These halogenated hydrocarbons have been shown to be highly effective in extinguishing fires in critical situations by virtue of their flame-inhibiting characteristics, nontoxicity prior to use, and relative nontoxicity during and following use, and because ancillary damage caused by their presence after extinguishment is minimized. Critical uses range from extinguishing fuel fires in military vehicles while preserving crew safety, to safe and nondamaging extinguishment of electrical fires in large computer facilities. Unfortunately, Halons 1301 and 1211 have been implicated as participants in stratospheric ozone depletion (Rowland 1991), and their production and sale have been banned by international agreement (Copenhagen 1992) as of 1 January 1994. Because of this, there is a concerted effort underway to find environmentally friendly yet effective substitutes for these fire inhibitors (Nyden et al. 1994) for which production has been halted.

There is a large body of literature devoted to the investigation of the mechanism by which certain additives, or diluents, affect combusting systems (Gann 1975; Bajpai 1974). We define a diluent as any dopant (i.e., inhibitor, suppressant, retardant, extinguisher, negative catalyst, accelerant, fuel/oxidizer added to a combusting system, etc.) to the existing flame system, regardless of the effect on the system. For investigations of flame inhibitor performance, scenarios range from studies of inhibitor effectiveness on large fires fueled by liquid hydrocarbons burning in air, to inhibitor effect on laminar low pressure premixed gas flames. Unfortunately, these studies have shown that the effectiveness of a particular fire inhibitor may vary depending on the type of combusting system being investigated (Biordi, Lazzara, and Papp 1975). For most real fires, the conditions within the combusting region may vary from highly diffusion-limited, fuel-rich combustion to locally premixed stoichiometric combustion. The inhibitor therefore needs to be effective over a wide range of macroscopic combustion parameters.

2. BACKGROUND

A macroscopic effect of adding a diluent to a premixed volume of gas through which a combustion wave is propagating is to alter the flame velocity (Lewis and von Elbe 1987b). The flame velocity is highly dependent on the fuel/oxidizer equivalence ratio, defined here for the uninhibited flame as the ratio of mixture volume percent fuel divided by mixture volume percent oxidizer to the stoichiometric balanced

chemical equation coefficient of fuel divided by the coefficient of oxidizer. For the flames studied here, this equation is $\text{CH}_4 + 2\text{O}_2 = \text{CO}_2 + 2\text{H}_2\text{O}$. By this convention, a fuel-rich flame will have an equivalence ratio greater than 1 and a fuel-lean flame will have an equivalence ratio less than 1. Depending on the equivalence ratio, and whether the diluent might be classed as inert, or an additional fuel or oxidizer, or an inhibitant, or any combination of these classifications, this change in flame velocity can be positive or negative. For the case of a diluent that may be classed as an inhibitant, the overall effect is to lower the flame velocity, with incremental addition of inhibitant eventually resulting in extinction of the flame.

All diluents will have an effect by virtue of the increase they bring to the heat capacity of the premixed gas (Williams 1965). The higher heat capacity of the mixture, upon addition of a diluent, initially removes energy from the flame. The energy removal lowers the flame temperature, reducing the rate of elementary chemical reactions responsible for flame propagation. This inhibitory effect is a physical manifestation. The secondary effect of the diluent is dependent upon whether the temperature and constituents within the flame enable the diluent to become a participant in the combustion chemistry. When the chemical reaction of the diluent with the premixed gas in the flame promotes flame velocity to the extent that the physical reduction of the flame velocity is overcome, the diluent may be classed as a fuel. When the diluent does not participate in the combustion chemistry, or when the diluent does participate and diluent-generated species interfere or suppress generation of species important to flame propagation, the diluent may be classed as an inhibitant.

Measurement of the speed of a laminar flame propagating through a premixed gas is most often accomplished using the total area method from a schlieren image of a flame supported by a Bunsen-type burner (Williams 1965). When operated in a vertical configuration, the premixed gas exits from the burner tube orifice and is confined from above by a stationary conical combustion wave. Except for the base and the tip of the cone, the wave is usually considered to be planar. Schlieren images of the height of the inner edge of the cone are used to geometrically calculate the area of the cone. The burning velocity is calculated by dividing the gas flow rate by the area of the combustion cone. Burning velocities calculated in this way for many different fuel/oxidizer systems can be a means of verifying flame kinetic mechanisms used in adiabatic laminar flame computer models (Linteris, to be published).

We report here the development of a new device that uses pulsed flame velocimetry (PFV) to measure the degree to which an inhibitor decreases flame speed. The device is based upon the pulsed flame

detection method developed at Tel Aviv University that was applied in a pulsed flame photometer detector by Cheskis, Atar, and Amirav (1993) and in a combined pulsed flame photometer and ionization detector as described by Tzanani and Amirav (1995). Results of flame velocity measurements using this device show good correlation with previous results of flame velocity measurements, and good correlation with previous results of extinguishment studies. The new device is simple, inexpensive to build and operate, and provides repeatable results. It is especially well suited to studies of flame inhibition by Halons because it relies on a nonoptical technique, minimizing equipment cost and potential damage to sensitive optics from exposure to hot acid gases.

3. EXPERIMENTAL

The pulsed flame velocimeter consists of a rectangular aluminum block, 10 cm in length with a square cross section of 25 cm² (see Figure 1). A 0.5-cm-diameter circular cross section cavity has been bored down the center and the length of the Al block. Bored into one side of the block and into the central cavity (normal to the cavity axis) are three 1-cm-diameter holes that house an igniter and two charge collectors. The igniter is positioned in the hole nearest the cavity exit. At the cavity end farthest away from the igniter (resistively heated Pt wire) is a plug with a small aperture allowing a flow of gas to pass into the cavity. BNC-type charge collectors, separated by 50 mm, are positioned in the remaining holes normal to the cavity axis. The entire apparatus is encased in heating tape and temperature is controlled and monitored (Omega Engineering) using Pt-Pt/10%Rh thermocouples embedded at opposite ends of the device. It is assumed that during operation, no temperature gradients exist within the aluminum block. Device temperature measured by thermocouples embedded in opposite ends of the device always agreed to within 2 K.

To operate the device, a premixed fuel/oxidizer/inhibitant gas is allowed to pass through the aperture into the cavity end furthest from the igniter. When the flowing column of gas reaches the heating element, the gas ignites and the combustion wave traverses the tube toward the rear of the device, passing the two charge collectors, which are connected in series. As the combustion wave passes each charge collector, a small voltage is generated, which is recorded on a digital oscilloscope (LeCroy Industries), enabling calculation of flame velocity. When the combustion wave reaches the gas entrance aperture, combustion is quenched by a flame arrestor placed in the mouth of the aperture. A fresh sample of gas continuously flows into the device, exhausting the burned gas in the cavity, and the process repeats itself (see Figure 2). Gas flow velocity is negligible compared to flame velocity. Each time the velocimeter

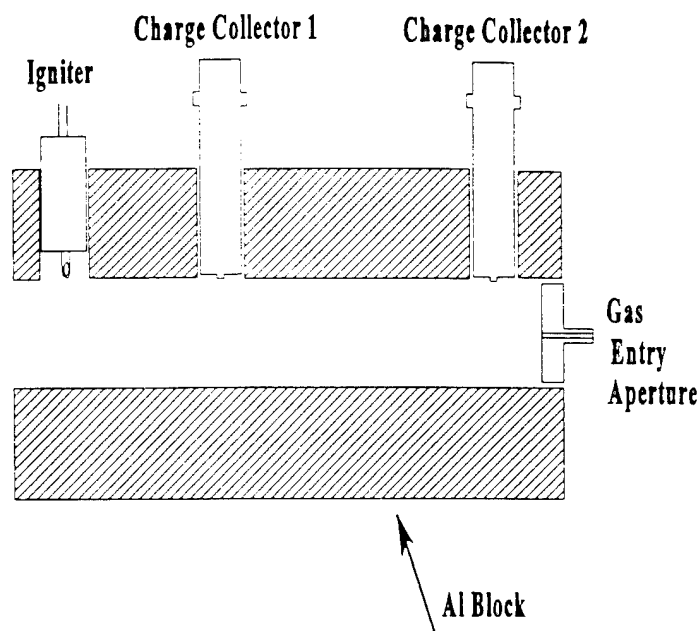


Figure 1. Cross-sectional view of the PFV showing the igniter, charge collectors, and gas entry aperture.

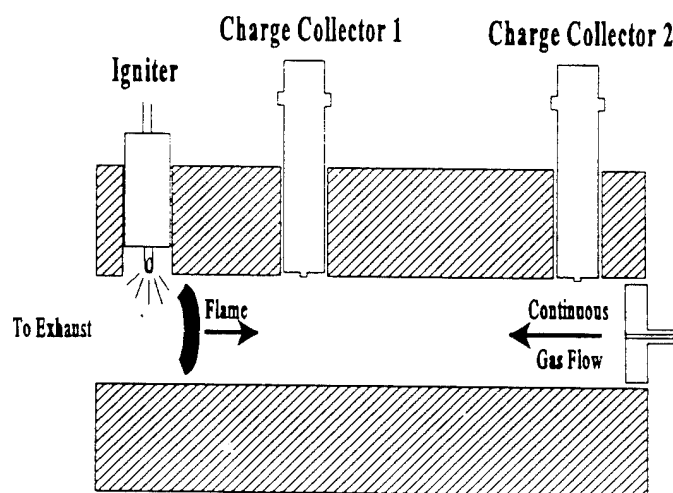


Figure 2. Cross-sectional view of the PFV during operation. The fuel/oxidizer gas mixture flows continuously. Gas flow rates are negligible compared to flame velocities.

operates, a small flame is observed at the open end of the device. Depending on device temperature, total flow rates varied from 0.214 to 0.3 liters/min, providing a flame repetition rate of approximately 2 Hz. Gas flow was controlled using calibrated gas flow controllers (MKS Industries). The fuel/oxidizer equivalence ratio found to give most repeatable operation was 0.35. At equivalence ratios higher (more rich) than this value, measured flame velocities showed increased scatter.

For methane/oxygen flames, steady operation at a total flow of 0.214 liters/min causes the device temperature to remain at approximately 355 K. For Halon-inhibited flames, lower temperatures of the pulsed flame caused this temperature to vary, depending on inhibitor. For this reason, and to minimize reactions of acid gases with condensed water vapor on the walls of the cavity, the device was heated to and maintained at a temperature of 385 K. The tradeoff for operation at this temperature was that the velocimeter would not function properly at this temperature at an equivalence ratio higher than 0.35. We believe that modification of the gas entry aperture (to minimize turbulence and increase quenching efficiency) will enable operation at a greater range of equivalence ratios.

The average value of the flow rate of the premixed gas through the velocimeter was $3.6 \text{ cm}^3\text{s}^{-1}$. This means that for a cavity length of 10 cm and a cavity diameter of 0.5 cm, the expected pulsed flame frequency is 1.8 Hz, and, considering the velocimeter temperature of 355 K (gas expansion), the pulsed flame frequency is expected to be 2.2 Hz. This value is in close agreement with the ~2 Hz operational frequency observed.

Gases were not preheated prior to entering the velocimeter. Each reported flame velocity represents the summed average of 100 repetitions of the device. Measured flame velocity in tubes has been shown to be angle sensitive (Lewis and von Elbe 1987a), so all data reported here were measured with the cavity tilted approximately 10° above horizontal, with the open end of the cavity directed slightly upward. It was found that for flames inhibited by Halons, generation of HF gas caused the Pt filament to burnout after only a few hundred repetitions. For the experiments using Halon inhibitors, the Pt igniter was replaced by a methane/air flame positioned approximately 1 mm away from the open end of the cavity.

All experiments were conducted at atmospheric pressure. Comparisons of flame speeds for uninhibited methane/oxygen flames showed the method of ignition to have a negligible effect, with the flame-ignited measurements yielding a slightly higher flame speed than the glow-ignited measurements, possibly because the flame at the cavity entrance caused excess heating of the gas just prior to flame ignition. No difference in flame velocities in neat methane/oxygen mixtures was seen when the Pt filament was exchanged for a Kanthal AF-1 wire filament or for a Nichrome filament. When the Pt filament was used, decreasing filament current after the device was warmed to operating temperature caused the device to stop operating, indicating that the Pt filament was not igniting the gas catalytically. All gases were supplied by Matheson, Inc., and were used without further purification.

4. RESULTS AND DISCUSSION

Figure 3 shows the measured flame velocity for uninhibited flames as a function of equivalence ratio for the pulsed flame velocimeter with the velocimeter temperature maintained at 355 K. Total flow rates varied from 0.06 to 0.3 liters/min. Above this temperature, the device would not operate at an equivalence ratio higher than 0.35. Above this equivalence ratio, the flame would stabilize at the aperture through which gases entered the cavity, negating any attempts to measure flame velocities at 385 K at an equivalence ratio higher than 0.35.

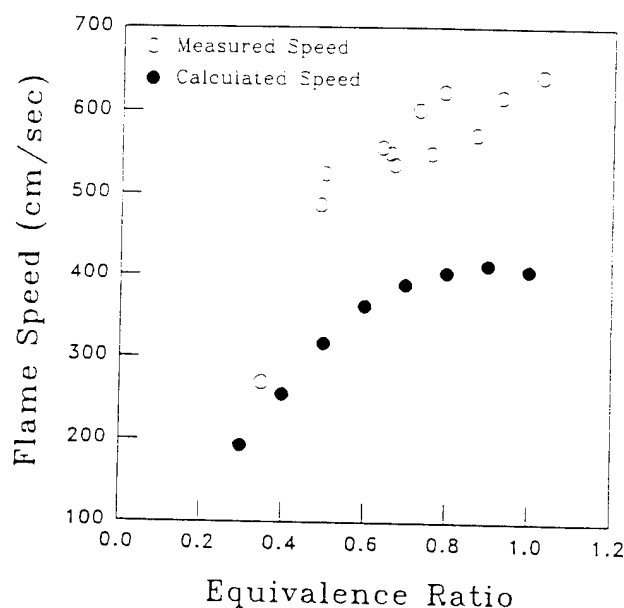


Figure 3. Flame velocity as a function of equivalence ratio measured with the PFV, and calculated using the SANDIA laminar flame code. The premixed gas temperature is 355 K.

Figure 3 shows flame velocities for methane/oxygen flames as a function of equivalence ratio measured with the PFV and calculated using the adiabatic laminar flame code for a premixed gas temperature of 355 K. The flame speed of a premixed methane/oxygen flame was calculated using computer codes developed at the Sandia National Laboratories. The flame speed is calculated for a freely propagating flame at an initial temperature of 355 K at 1.0 atmosphere pressure. For the calculations, the GRIMECH12 kinetic mechanism of methane combustion was employed. For each equivalence ratio in Figure 3, calculated laminar adiabatic flame velocities are less than what is measured using the PFV (Kee et al. 1985).

Figure 4 shows a drawing of the appearance of the flame front in a horizontally positioned tube containing a premixed flammable gas when the tube is closed at one end and ignited at the open end (Zeldovich 1981). Gas flow near the walls of the cavity is slower than near the axis of the cavity because of viscous drag. Because expanding gas cannot escape from the closed end of the tube, gas flows opposite to the direction of the combustion wave near the walls of the cavity. Buoyancy of the hot gases within the tube creates the asymmetry of the combustion wave. The flame velocity relative to the tube is equal to the sum of the normal flame velocity and the horizontal component of the cold gas velocity near the most advanced flame point in Figure 4. The flame velocity measured by the velocimeter (the velocity relative to the cavity) may be given by (Zeldovich 1981):

$$V = AV_{\text{lam}}/\pi R^2$$

where V is the measured flame velocity, V_{lam} is the calculated laminar adiabatic flame speed, A is the area of the flame surface, and R is the radius of the cross section of the cavity. While not accounting for heat losses to the walls of the device as a function of the flame surface, or for the asymmetry of the flame front with regard to the cavity axis, this equation does show how measured velocities in the PFV may be higher than those calculated using adiabatic flame codes. We believe the scatter in the experimental data in Figure 3 at high flame velocity (high equivalence ratio) is caused by turbulence due to elongation and asymmetry of the flame front.

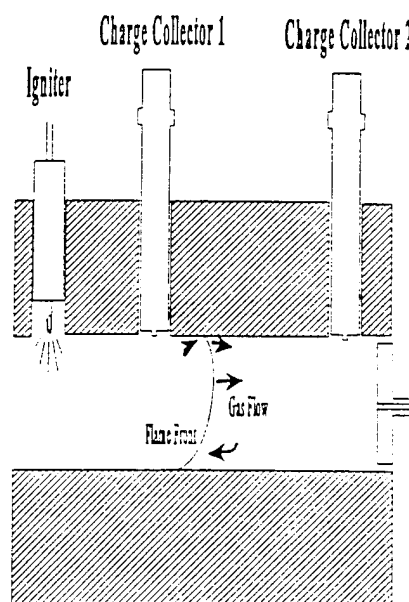


Figure 4. Approximate appearance of the flame front and gas motion in a horizontal tube closed at one end and containing a premixed fuel/oxidizer gas after ignition at the open end. Note gas buoyancy effects and asymmetry of flame front with respect to the cavity axis. This figure is adapted from Zeldovich (1981).

Figure 5 shows measured flame velocities for a premixed methane/oxygen flame inhibited by CF_3Br as a function of percent volume CF_3Br at two different device temperatures. From this figure, it may be seen that heat transfer from the walls of the cavity to the premixed gas has an important effect on measured flame speed. The slight sigmoid shape of the data at low levels of inhibitor concentration is common to all data measured using the device. At this time, we do not have an explanation for this behavior. We estimate the error in flame speeds measured with a fuel/oxidizer equivalence ratio of 0.35 to be approximately 10%.

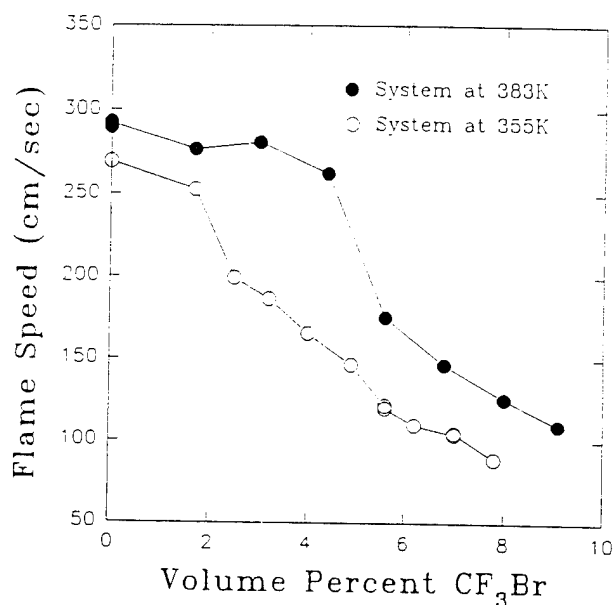
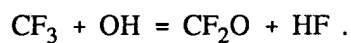


Figure 5. Flame speeds of methane/oxygen flames ($\phi = 0.35$) inhibited by CF_3Br with the PFV temperature maintained at 355 K and 383 K.

From the previous discussion, it may be argued that the velocimeter measurements differ from total area method measurements in several ways. By the nature of the design, the premixed gas is mixed with burned gas prior to reaching the igniter. Because the igniter represents a constant energy source, the premixed gas will dilute the burned gas in the cavity until a flammability limit is reached, at which time the velocimeter will operate. Recently, some experiments were conducted in which flame velocity was measured as a function of distance from the igniter, using a velocimeter equipped with optical ports through which OH emission (from the flame front) was monitored. It was found that over the length of the device, the flame velocity was constant to within a few percent. Since this would not be the case if mixing of the premixture with burned gases were important, we have concluded that this is not a major source of error in the data. Also, it has been shown that the temperature of the walls of the cavity play

a significant role in determining flame velocity, possibly because the temperature influences the asymmetry of the flame front, or influences heterogeneous chemistry occurring on the tube walls.

Figure 6 shows flame speeds of premixed methane/oxygen flames inhibited by Ar, CF_2H_2 , CF_3H , and CF_4 as a function of volume percent inhibitor. Inhibition of flames by Ar is assumed to occur via a purely physical mechanism. The effect of addition of CF_2H_2 to the premixed gas is to act as a fuel. After only a few percent addition of CF_2H_2 , the effective equivalence ratio of the premixed gas was raised beyond 0.35 to the extent that the velocimeter ceased to operate, with the $\text{CH}_4/\text{O}_2/\text{CF}_2\text{H}_2$ flame burning at the aperture entrance to the cavity. CF_3H was found to be superior to Ar as a flame speed inhibitor on a percent volume basis, and CF_4 was found to be slightly superior as a flame speed inhibitor to CF_3H . This result is in contrast to laminar flame speed calculations (Linteris, to be published) and to measurements of flame speeds using the total area method. Laminar adiabatic flame modeling calculations predict that CF_4 acts as an inhibitor only by changing the heat capacity of the combusting mixture. CF_3H , however, is a good source of the CF_3 radical, which may react via the mechanism (Vandooren, Nelson da Cruz, and Van Tiggelen 1988):



This reaction acts as an OH scavenger, so CF_3H is expected to inhibit combustion via a chemical mechanism. The discrepancy between the PFV measurements and total area method measurements of flame speeds for CF_3H inhibited flames relative to CF_4 inhibited flames may be caused by the lower gas temperatures present in the PFV combustion wave.

Figure 7 shows flame speeds of the inhibited methane/oxygen flame ($\phi = 0.35$) as a function of weight percent inhibitor. It has been suggested (Larsen 1975) that flammabilities of fuel/oxidizer/halocarbon gas mixtures are dependent on the weight percent of halocarbon in the mixture. From Figure 7, it may be seen that when flame speed reduction is plotted vs. weight percent inhibitor, inhibition efficiency among the fluorocarbons becomes more equal than when expressed on a volume percent basis. This indicates that flame speed reduction via a physical mechanism is important when flame inhibition is measured using the PFV. The fact that CF_2H_2 acts as a fuel indicates that the device is also sensitive to diluents which may participate in flame chemistry. Also, it may be seen that on a weight percent basis up to approximately 20%, CF_3H and CF_4 cause similar flame speed reductions.

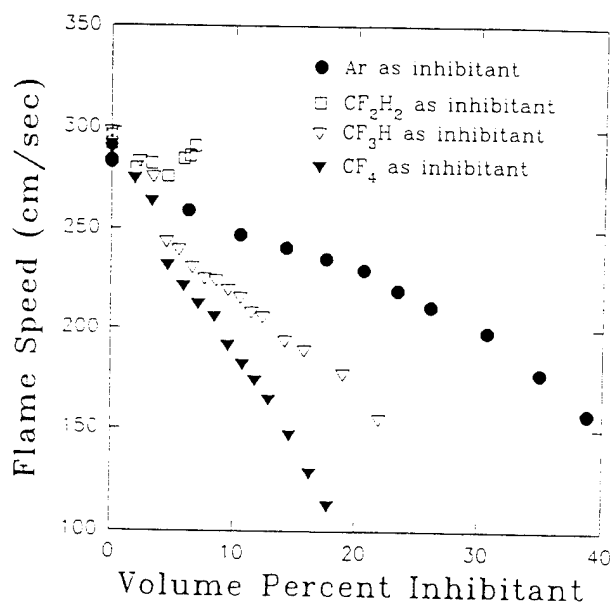


Figure 6. Flame speeds of methane/oxygen flames ($\phi = 0.35$) inhibited by Ar, CF₂H₂, CF₃H, and CF₄ as a function of volume percent inhibition.

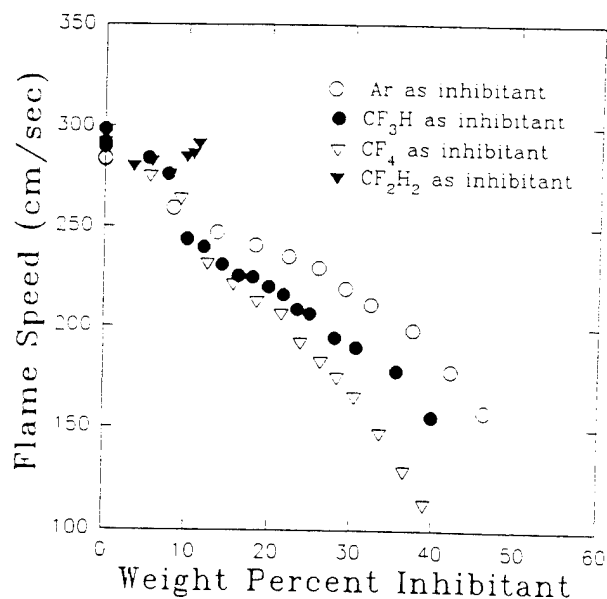


Figure 7. Flame speeds of methane/oxygen flames ($\phi = 0.35$) inhibited by Ar, CF₂H₂, CF₃H, and CF₄ as a function of weight percent inhibitor.

Figure 8 shows flame speeds of the methane/oxygen flame ($\phi = 0.35$) inhibited by Ar, CF₄, and by CF₃Br (Halon 1301) as a function of volume percent inhibitor. Of all inhibitors examined in this study, CF₃Br was the most effective on a percent volume basis in reducing flame velocity. Figure 9 shows the data in Figure 8 plotted on a weight percent basis. From this figure, it may be seen that CF₃Br is still the "best" flame speed reducer examined in this study, but that the difference in flame speed reduction between methane/oxygen flames ($\phi = 0.35$) doped with CF₃Br or CF₄ when examined on an inhibitor weight percent basis is less than what is observed when the flame speed reduction is examined on an inhibitor volume percent basis.

5. CONCLUSION

Using the pulsed flame velocimeter, it is possible to measure the effect of various diluents on flame speed in a flame propagating in a tube containing a premixed methane/oxygen gas mixture. Results of measurements using the pulsed flame velocimeter of flame speed reduction in methane/oxygen mixtures by various Halons are qualitatively similar to measurements of flame speed obtained using the total area method. The device holds promise as a fast, efficient, and inexpensive tool by which flame inhibitor efficiency may be measured. However, more measurements on a wide variety of inhibitors need to be

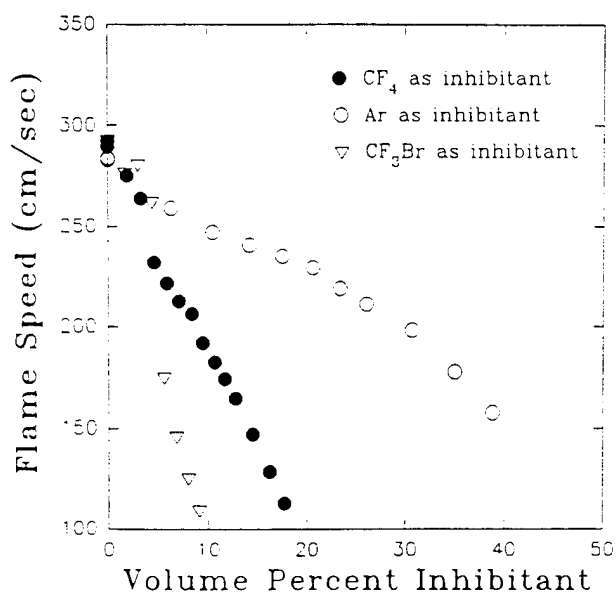


Figure 8. Flame speeds of methane/oxygen flames ($\phi = 0.35$) inhibited by Ar, CF₄, and CF₃Br as a function of volume percent inhibitor.

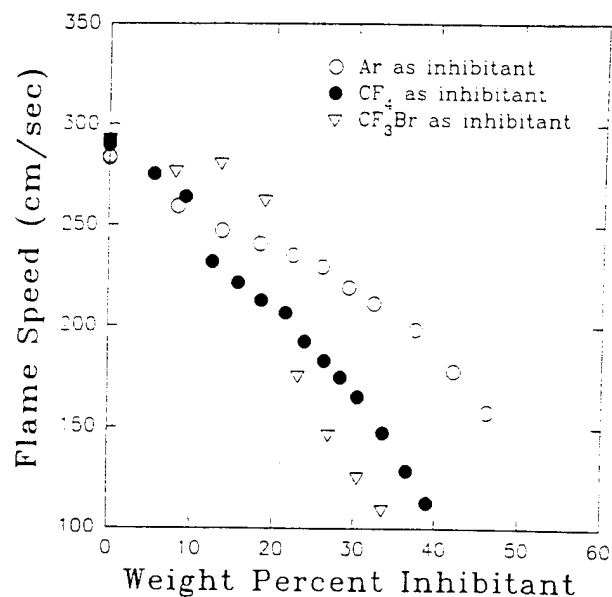


Figure 9. Flame speeds of methane/oxygen flames ($\phi = 0.35$) inhibited by Ar, CF₄, and CF₃Br as a function of weight percent inhibitor.

done before the device can become a compliment to the many other test methods available for predicting flame inhibitor efficiency. Measurements are currently under way using a new series of inhibitors, and the device has been redesigned to allow operation at higher fuel/oxidizer equivalence ratios.

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6. REFERENCES

- Bajpai, S. N. "An Investigation of the Extinction of Diffusion Flames by Halons." Journal of Fire and Flammability, vol. 5, pp. 255-267, 1974.
- Biordi, J. C., C. P. Lazzara, and J. F. Papp. "The Effect of CF_3Br on Radical Concentration Profiles in Methane Flames." Halogenated Fire Suppressants, edited by R. G. Gann, ACS Symposium Series No. 16, The American Chemical Society, pp. 256-294, Washington, DC, 1975.
- Bowman, C. T., M. Frenclach, W. C. Gardiner, D. Golden, V. Lissianski, G. Smith, and H. Wang. "GRIMECH," upcoming Gas Institute report, 1995.
- Cheskis, S., E. Atar, and A. Amirav, A. "Pulsed Flame Photometer: A Novel Gas Chromatography Detector." Anal. Chem., vol. 65, pp. 539-555, 1993.
- Copenhagen revisions to "The Montreal Protocol on Substances That Deplete the Ozone Layer," (1987) 1992.
- Gann, R. G. (ed.). "Halogenated Fire Suppressants." ACS Symposium Series No. 16, The American Chemical Society, Washington, DC, 1975.
- Kee, R. J., J. F. Grcar, M. D. Smooke, and J. A. Miller. "A Fortran Program for Modeling Steady Laminar One-Dimensional Premixed Flames." Sandia National Laboratories Report, SAND85-8240, 1985.
- Larsen, E. R. "Halogenated Fire Extinguishants: Flame Suppression by a Physical Mechanism?" Halogenated Fire Suppressants, edited by R. G. Gann, ACS Symposium Series No. 16, The American Chemical Society, Washington, DC, pp. 376-402, 1975.
- Lewis, B., and G. von Elbe. Combustion, Flames, and Explosions of Gases. New York: Academic Press, Inc., Third Edition, pp. 305-315, 1987a.
- Lewis, B., and G. von Elbe. Combustion, Flames, and Explosions of Gases. New York: Academic Press, Inc., Third Edition, pp. 395-417, 1987b.
- Linteris, G. T. "The Effect of Inhibitor Concentration on the Inhibition Mechanism of Fluoromethanes in Premixed Methane-Air Flames." To be published in upcoming ACS Symposium Series on Halon Alternatives.
- Nyden, M. R., G. T. Linteris, D. R. F. Burgess, Jr., P. R. Westmoreland, W. Tsang, and M. R. Zachariah. "Flame Inhibition Chemistry and the Search for Additional Fire Fighting Chemicals." Evaluation of Alternative In-Flight and Dry Bays, (Eds. W. L. Grosshandler, R. G. Gann, and W. M. Pitts), NIST SP 861, 467-641, National Institute of Standards and Technology, Gaithersburg, MD, 1994.
- Rowland, F. S. "Stratospheric Ozone Depletion." Annual Reviews of Physical Chemistry, vol. 42, pp. 731-768, 1991.

- Tzanani, N., and A. Amirav. "The Combined Pulsed Flame Photometric Ionization Detector." Anal. Chem., vol. 67, pp. 167–173, 1995.
- Vandooren, J., F. Nelson da Cruz, and P. J. Van Tiggelen. "The Inhibiting Effect of CF_3H on the Structure of a Stoichiometric $\text{H}_2/\text{CO}/\text{O}_2/\text{Ar}$ Flame." Twenty-Second Symposium (International) on Combustion, The Combustion Institute, pp. 1587–1595, 1988.
- Williams, F. A. Combustion Theory, Addison-Wesley Publishing Co., pp. 95–134, Reading, MA, 1965.
- Zeldovich, Y. B. "Structure and Stability of Steady Laminar Flame at Moderately Large Reynolds Numbers." Combustion and Flame, vol. 40, pp. 225–234, 1981.

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